

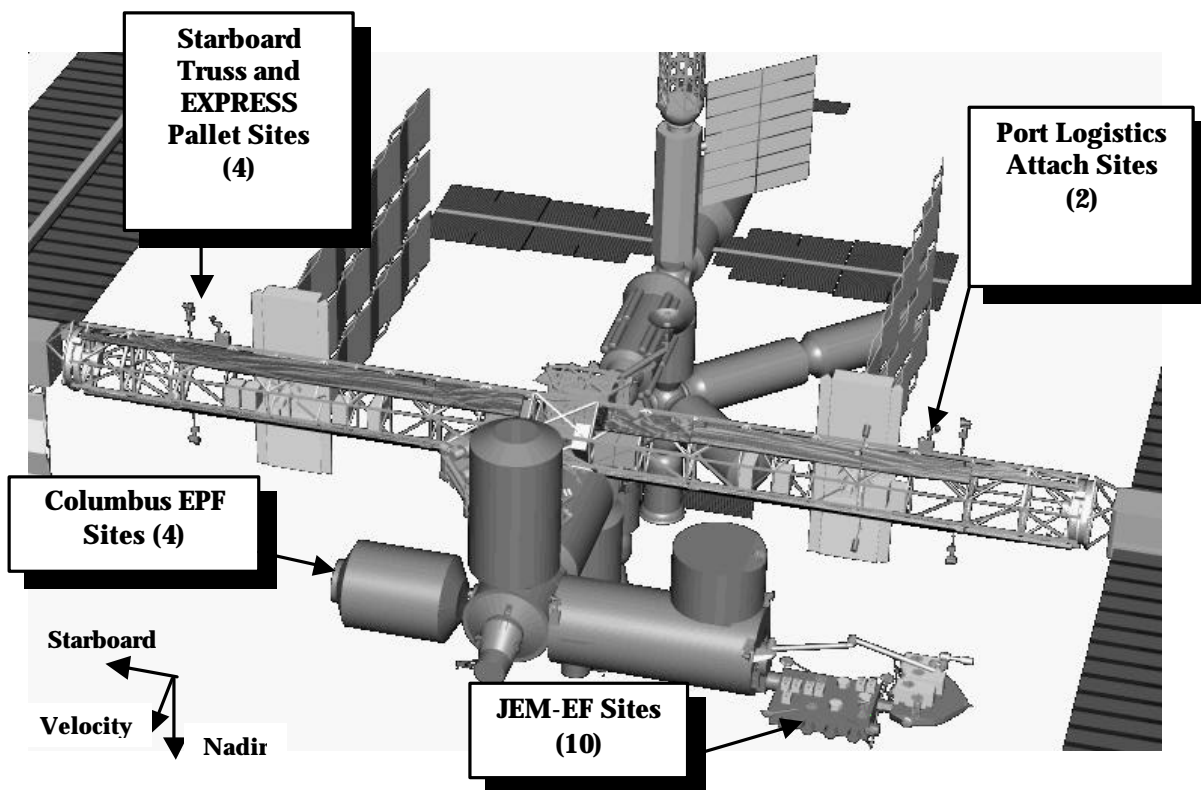
# INTERNATIONAL SPACE STATION (ISS) RESEARCH OPPORTUNITIES

December 12, 2002

## SECTION 1: INTRODUCTION

The International Space Station will provide opportunities for attached payloads at several external locations. These locations consist of the U.S. Truss, the U.S. Truss via the Expedite the Processing of Experiments to Space Station (EXPRESS) Pallet, the Japanese Experiment Module Exposed Facility (JEM-EF), and the Columbus External Payload Facility (EPF). Each site offers unique capabilities and environments. Non-standard sites, which may not provide power or data through the ISS, may also be considered on a case-by-case basis. In addition to the attached payload opportunities, additional research opportunities are available in the internal, pressurized Window Observational Research Facility (WORF). Refer to the NASA Announcement of Opportunity for current International Space Station payload attach sites being offered.

Figure 1 shows the standard attached payload and logistics sites on the ISS. The WORF, which resides in the window in the U.S. Lab, is not visible in this view.



*Figure 1. ISS External Payload Accommodations*

## SECTION 2: ISS ENVIRONMENT

The ISS is currently in the process of being assembled on-orbit. While assembly is not yet complete, attached payloads will be flown during assembly after the attach sites become available. Payloads for the U.S. Truss sites are anticipated to be manifested no earlier than 2005, followed by the JEM-EF in 2007. At this time, there is no opportunity for attached payloads to return on the Shuttle until after assembly complete.

The WORF is scheduled for launch in 2003. WORF payloads may be launched on subsequent flights. Launch and return manifesting and on-orbit duration will be determined for individual payloads based on payload requirements and Shuttle pressurized space availability.

## Orbit

The ISS orbit will have an inclination of 51.6 degrees with an altitude that varies between 350 and 470 Km due to the solar cycle. After initial assembly operations, the ISS orientation is continuously pointed nadir to earth. The potential viewing zone covers approximately 85% of the globe and 95% of the Earth's population. The orbit regression rate is one full orbit in two months and the ISS will fly over the same spot on the Earth's surface approximately every 3 days. It will cover the same spot with the same lighting conditions approximately every 3 months. The ISS will pass regularly through the South Atlantic Anomaly.

## Attitude

The ISS will use the Global Positioning System (GPS) to determine the ISS state vector (position and velocity), attitude and altitude rates, and a time reference. GPS antennas will be located at the S0 truss segment. The system will provide a total position error of <3000 feet RSS (3 sigma) and an attitude error of <0.5° per axis (3 sigma) at the GPS antenna site. This position knowledge will degrade with distance from the GPS antenna site with a worst-case knowledge of 3.0 degrees (3 sigma) and an expected knowledge of ~1 – 2 degrees (3 sigma) at the S3 truss attach sites. The Station will estimate the on-orbit inertial rates at the GPS origin to within 0.01° per second per axis, at a 0.5 Hz bandwidth. More information on pointing can be obtained from the "Space Station GN&C Overview for Payloads" on the Research Program Office website at <http://rpo-iss.gsfc.nasa.gov>.

Maximum excursions are specified to be +/- 15 degrees in roll and yaw, and +15 to -20 degrees in pitch with respect to the Local Vertical Local Horizontal (LVLH) attitude. During the periodic reboost of ISS, the station will maneuver 180 degrees in yaw prior to reboost, then maneuver back. The vehicle stability will be at least 3.5 degrees/axis/orbit (peak-to-peak) and it is expected to be better than 2.0 degrees/axis/orbit (peak-to-peak). Payloads may need an active pointing system and star tracker to accurately point at chosen targets. Pointers will not be provided by the ISS program and are the sole responsibility of the Payload Developer. Pointers are being developed by European Space Agency (ESA) for specific EXPRESS Pallet adapter payloads.

## ISS Induced Environment

The requirements for the overall ISS induced environment specify the values for molecular column density and permanent molecular deposition as seen in Table 1.

Table 1. Overall ISS Induced Environment Requirements

Parameter Modeled	ISS Requirements: Quiescent	ISS Requirements: Non-quiescent
Molecular Column Density	$1 \times 10^{14}$ molecules/cm <sup>2</sup>	Unlimited density is allowed, but frequency is limited
Permanent Molecular Deposition	$1 \times 10^{-14}$ g/cm <sup>2</sup> s <sup>-1</sup> (-30 Å/yr)	$1 \times 10^{-6}$ g/cm <sup>2</sup> s <sup>-1</sup> (-100 Å/yr)

There are four primary sources of induced contamination: deposition due to material outgassing, water dumps, CO2 dumps, and waste gas vents from pressurized payload racks or other attached payloads. Outgassing deposition at the attached payload sites is due primarily to neighboring attached payloads. Waste gas vents are short in duration and should be scheduled. CO2 vents cannot be scheduled, but they do not exceed ISS induced environment requirements. Water dumps are scheduled events.

The ISS Program and the appropriate Research Program Offices are developing molecular deposition requirements for individual payloads. Payloads that are extremely contamination sensitive should consider making their own deposition measurements in addition to providing protection for contamination sensitive surfaces. The deposition contamination requirements for the full truss sites, which include the integrated pallets, restrict molecular deposition to  $1 \times 10^{-15}$  g/cm<sup>2</sup> s on sensitive ISS elements and  $1 \times 10^{-14}$  g/cm<sup>2</sup> s on other payloads. The contamination environment of the EXPRESS Pallet location will necessarily be determined by the Pallet analytical integration once all Pallet payloads have been selected. Contamination requirements for EXPRESS pallet payloads are located in the EXPRESS Pallet Interface Definition Document (SSP 52000-IDD-EPP). The payload contamination requirements for the JEM-EF and Columbus EPF sites have not yet been determined.

## Safety

System safety from both a Space Shuttle and ISS standpoint will require significant consideration. The safety process is described in NSTS 1700.7, Safety Policy and Requirements for Payloads using the Space Transportation System and ISS Addendum and NSTS 13830, Implementation Program for STS System Safety Requirements.

Payloads will be required to complete the safety review process prior to shipping their equipment to Kennedy Space Center (KSC) for integration on the ISS carrier hardware and installation into the shuttle. Upon completion of on-orbit activities on the ISS, all U.S. payloads are required to be returned to earth via the shuttle. Payloads are therefore required to address the retrieval hazards in their flight hardware designs and during the pre-launch safety review process. Prior to the actual retrieval operations, the payload organization must support a delta safety review.

## Launch Vehicle

Full truss site, EXPRESS Pallet, JEM-EF, Columbus EPF, and WOLF payloads will be launched on the Space Shuttle. With the exception of full truss payloads, carriers for individual payloads will be provided by the ISS Program. JEM-EF payloads may also be launched on the Japanese HTV vehicle, although they must return on the shuttle.

## Fields of View

Fields of view for some attached payload sites are available on the RPO website.

### **SECTION 3: FULL TRUSS SITE PAYLOAD ACCOMMODATIONS**

The U.S. Truss attached location provides the capability for large payloads to be mounted to the S3 site. Currently, the programmatic planning utilizes one zenith site for a single large payload while the other three sites are planned for EXPRESS Pallet payloads. Figure 2 depicts a full truss attached payload. Use of this site requires the payload developer to provide all attach mounting hardware for both the S3 site as well as installation and mounting hardware in the shuttle.

#### Capability Per Payload

The payload mass is limited by the position of the payload center of gravity (CG) relative to the Payload Attach System (PAS). For a payload with a centered CG falling within the height restrictions, the mass limit is:

- 1360 Kg (3,000 lb) for a CG with X +/- 32 in, Y +/- 32 in, Z between 0 and +100 in
- 8618 Kg (19,000 lb) for a CG with X +/- 32 in, Y +/- 32 in and Z between 0 and +66 in.

The payload envelope is 2.23m along the truss and 5.0m in the ram/wake direction. The payload height limit is 3.0m plus 1.39m from the PAS interface to allow for trapezoidal support structure underneath. Payload height and width is also restricted by Shuttle bay envelope limits. The envelope is cutout near the PAS to maintain Extra Vehicular Activity (EVA) and Extra Vehicular Robotics (EVR) translation corridors between the payloads and the truss.

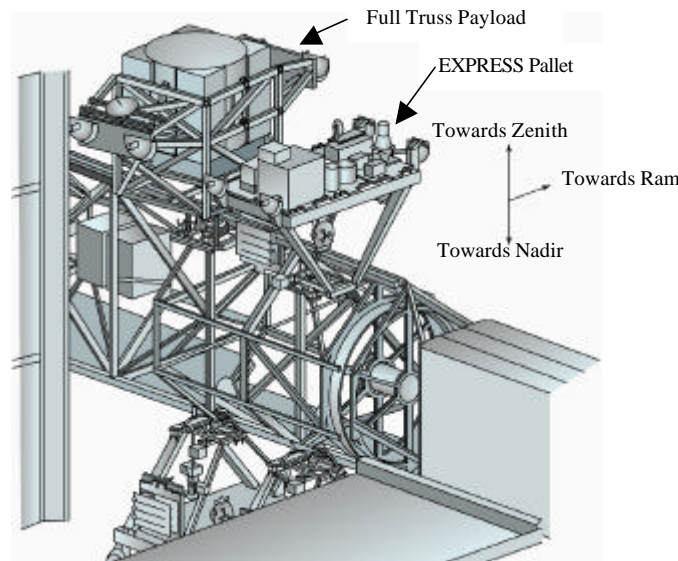
The maximum power capability of a full truss site is 3 kW. Due to power sharing among the attached payloads, an individual payload's power allocation is expected to be less than the full capability. Available power will drop to keep alive power levels during time periods in which the solar arrays are rotated off nominal to accommodate ISS vehicle, visiting vehicle, and crew activities. These activities include shuttle dockings and proximity operations as well as contingency modes. The keep-alive power allocation has not yet been determined. Payloads must provide their own thermal control during all power modes.

Low rate commanding and telemetry are available via the MIL-STD-1553B payload data bus for rates less than 20 Kbps. Fiber optics will provide a high rate data capability. Payloads should assume an average data generation rate of no more than 1-2 Mbps for the high rate data. The total downlink capability of the ISS Ku-band system is about 43 Mbps, but may be upgraded to 128 Mbps before assembly complete. High rate data downlink can be provided in near real time though it is expected that most external payloads will usually operate in a store and dump mode. A communications outage recorder capability will be provided to cover loss of signal periods, but not for nominal payload store and dump. It is recommended that payloads have their own data storage capability so that they can continue to operate when the shared data buses are not available or are in use by other payloads.

#### Transport and Installation

Full truss payloads will be launched and retrieved via the shuttle. The ISS Program is planning on supplying carriers for payloads up to 2500 Kg. Additional carrier interface hardware to use these carriers is the responsibility of the full truss payload. Payloads exceeding this mass must provide their own carrier interface to the shuttle. On orbit, the payload will be robotically removed from the shuttle and placed on the attach site. Retrieval will also take place robotically. Grapple

fixtures for the robotic arms are purchased by the payload from NASA. The berthing camera system necessary for robotic placement will be provided to the payload. EVA activity should be reserved for contingency operations only.



**Figure 2. Full Truss Payload**

#### **SECTION 4: EXPRESS PALLET PAYLOAD ACCOMMODATIONS**

The EXPRESS Pallet provides the capability for six payloads to reside on a full truss attach site as seen in Figure 3. Each payload sits on an adapter plate, which attaches and detaches from the EXPRESS Pallet.

##### Capability Per Payload

The payload mass limit is 227 Kg and must fit within an envelope of 1.1m (l) x 0.86m (w) x 1.2m (h). Power is provided via a 120 Vdc feed and a 28 Vdc feed. The maximum power available for an integrated pallet is 2.5 kW of combined 120 Vdc and 28 Vdc to be distributed between the six pallet adapter payloads. Each adapter site has the capability to provide 750 W of 120 Vdc and 500 W of 28 Vdc. In reality, power usage at each adapter site will probably be limited by the thermal dissipation capability, which is site dependent. No thermal control is provided by the Pallet, and dissipation of heat into the Pallet structure is limited to 50W. Keep-alive power during reduced power modes and contingency power for Pallet controller failures are provided at 120 Vdc only.

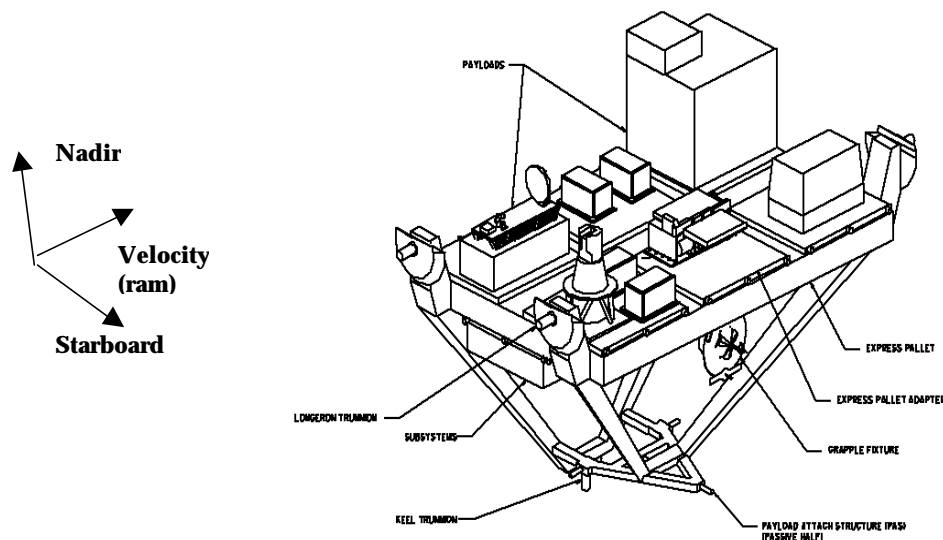
Data and command are available via a MIL-STD-1553B bus with an assumed maximum allocation of 20 Kbps to the integrated Pallet. Given the requirements of the pallet controller, each adapter payload can assume a 2 Kbps allocation. There are also 6 analog signals and 6 bi-level discretes available to each adapter payload, which are monitored by the pallet controller. Ethernet access to the high rate data link is available at ~6Mbps for the integrated pallet. Given the multiplexing of payload sites data through the high rate data link, a 250 Kbps average data rate should be assumed for individual Pallet payloads. Payloads will generally need to store and dump their high rate data. Data dumps may be transmitted at up to 6 Mbps burst rate or at an average rate of 1 Mbps depending on the operational scenario of each Pallet payload complement.

The six payloads on a nadir/zenith Pallet sit on the nadir/zenith face of the Pallet. The Pallet itself sits offset on the wake side of the U.S. Truss with its longest dimension perpendicular to the Truss. All six payloads have a nadir or zenith field-of-view. The two payloads closest to the Truss will also have a ram (ISS velocity direction) field-of-view. The two payloads farthest from the truss have a wake field-of-view.

##### Transport and Installation

EXPRESS Pallet payloads will be provided with a Pallet adapter plate, which also contains the necessary interfaces to the Station robotic arm. The first complement of Pallet payloads will be integrated with the EXPRESS Pallets at the Kennedy Space Center and launched on the EXPRESS Pallets. The integrated EXPRESS Pallet will be robotically installed on a Starboard attach site. At the end of their mission life, payloads on their Pallet adapter plate will be retrieved individually by

the Station robotic arm, placed on a carrier provided by the ISS Program, and returned via the Shuttle. As for the full truss payloads, EVA operations should be reserved for contingency only.



*Figure 3. EXPRESS Pallet configuration*

## SECTION 5: JAPANESE EXPERIMENT MODULE - EXPOSED FACILITY (JEM-EF) PAYLOAD ACCOMMODATIONS

The JEM-EF is attached to the port end of the Japanese pressurized module. This facility provides twelve attach sites: four on the ram side, four on the wake side, two on the port end, and two on the zenith face. One ram site and one port site are reserved for NASDA facility hardware; the Inter-orbit Communication System (ICS) and the Experiment Logistics Module–Exposed Section (ELM-ES). Of the remaining ten sites, five are allocated to NASA and five to NASDA. The JEM-EF configuration is shown in Figure 4. The sites on the side of the JEM-EF can all view zenith and nadir simultaneously. The two sites on the zenith face can only view zenith, ram, and wake.

### Capability Per Payload

The JEM-EF payload sites can accommodate two mass classes of payloads. Eight sites are limited to 500 Kg. The ram-port site, EFU #9, and the innermost wake site, EFU #2, can both accommodate 2500 Kg. However, due to mass limitations of the JEM Arm, issues regarding transport and installation of JEM-EF payloads >500 Kg exist. Contact the OES/OSS Research Program Office (Section 12) for more details. The payload envelope is 0.8 m x 1.0 m x 1.85 m with the longest dimension radiating from the JEM-EF.

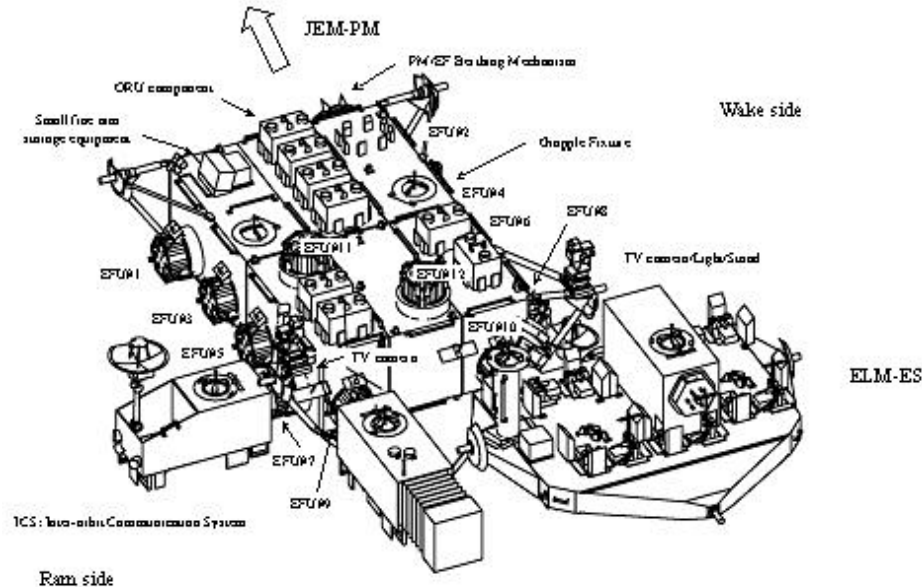
Power, data, and thermal accommodations on the JEM-EF are site dependent. The JEM-EF can provide a maximum of 10 kW of 120 Vdc power to all payloads simultaneously. Each site has the capability for 3 kW and the innermost ram and wake sites (EFU #1 and EFU #2) can provide 6 kW. Keep-alive power for each site is 100 W of 120 Vdc power. All payload sites are provided with active thermal control using coolant fluid loops. Each site has a maximum heat rejection capability of 3 kW except the innermost side sites, which can reject 6 kW. The maximum simultaneous heat rejection capability of the JEM-EF is 10 kW for payloads, which must be operationally shared with other external payloads and at some level with JEM internal payloads.

All sites provide a MIL-STD 1553B primary and secondary channel data interface. The eight payload sites on the sides of the JEM-EF connect to an FDDI 100 Mbps high rate data transmission line and a video line. Seven sites connect to the medium rate Ethernet transmission system. The MIL-STD 1553 and Ethernet connections can be configured to use either

the US or the NASDA data busses/LANs. Two of these sites are on the zenith face and the other five are on the sides. JEM-EF payloads should also plan to store and dump their data.

### Transport and Installation

Payloads may be launched via the Japanese H-IIA Transfer Vehicle (HTV) or the NASA shuttle. They will be installed robotically on the JEM-EF site. The JEM has its own robotic arm, which can service any JEM-EF payload. Payloads can also be accessed by the astronaut crew via the JEM airlock. The JEM arm can robotically access payload replacement hardware from the JEM airlock, but full payloads will not fit within the airlock volume. Payloads will be returned on the shuttle or may be destroyed through re-entry in the atmosphere by returning on the HTV.



**Figure 4. JEM-EF Configuration and JEM-EF Payload Attach Locations**

## **SECTION 6: COLUMBUS EXPOSED PAYLOAD FACILITY (EPF) PAYLOAD ACCOMMODATIONS**

The Columbus External Payload Facility (EPF) is located at the starboard end of the European Space Agency (ESA) Columbus Module. There are four payload attach sites, two of which are allocated to NASA and two to ESA. Two of the sites have nadir viewing, two have zenith viewing, and all four can view ram and limited wake. The Columbus with the EPF is shown in Figure 5.

### Capability Per Payload

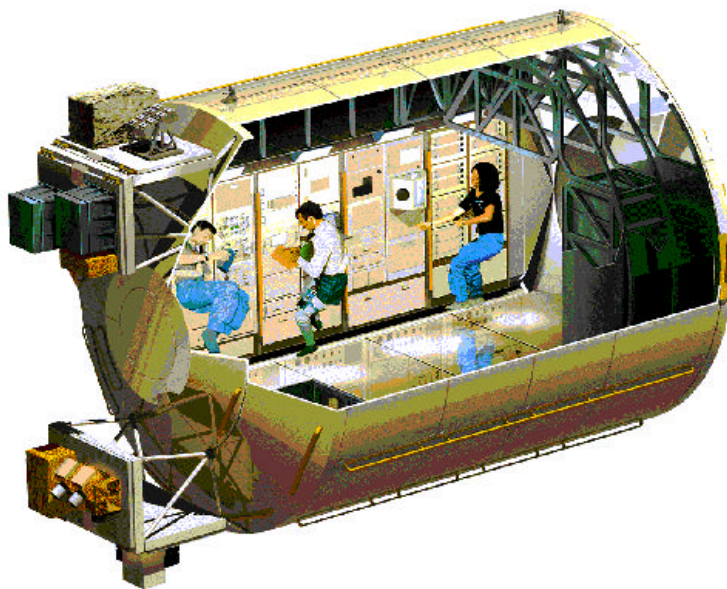
The EPF sites are being designed to accommodate an EXPRESS pallet adapter. The allowed mass and size will therefore be very similar to those for a subpallet payload; 227 Kg mass and a 1.1m (l) x 0.86m (w) x 1.2m (h) envelope.

The payload mechanical interface to the site is the same Flight Releasable Attachment Mechanism (FRAM) that is used to attach EXPRESS pallet adapters to the pallet. Payloads requiring a larger mass allocation may elect to use a different interface mechanism and will be considered on a case-by-case basis. Such payloads would be required to provide their own interface mechanism.

The payload power and data are provided through the Columbus pressurized services. Two redundant 120 V power feeds are provided per payload site. The EPF can provide 2.5 kW to any individual site, but the total power available for all four sites is also 2.5 kW, necessitating power sharing and lower power allocations per payload. Redundant MIL-STD-1553B data interfaces are provided for each site and can be configured to use either the U. S. 1553B payload data bus or the ESA 1553B payload bus. The Columbus also provides a redundant pair of LAN interfaces; again either U.S. or ESA LAN may be selected. The EPF will not provide active thermal control for any payload site.

#### Transport and Installation

EPF payloads will be transported to and from orbit on the NASA shuttle using an ISS program supplied carrier. Placement and retrieval will be accomplished via the shuttle and station robotic arms.



*Figure 5. Columbus Exposed Payload Facility Configuration*

### **SECTION 7: WINDOW OBSERVATIONAL RESEARCH FACILITY (WORF)**

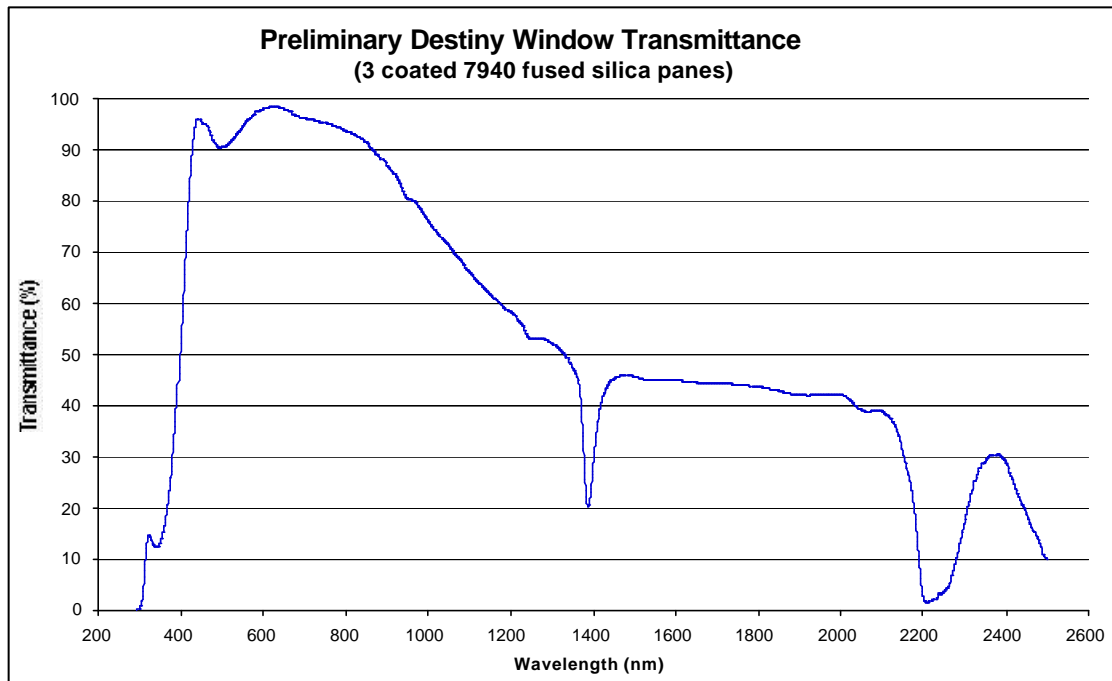
The Window Observational Research Facility (WORF) is a single-rack facility on ISS built to take advantage of the high optical quality nadir research window in the U.S. Laboratory.

#### Capability Per Payload

The WORF allows deployment of payloads as large as a 23 cm film aerial photography camera; the optical quality of the window allows deployment of payloads with optical diameters of up to 30.4 cm.

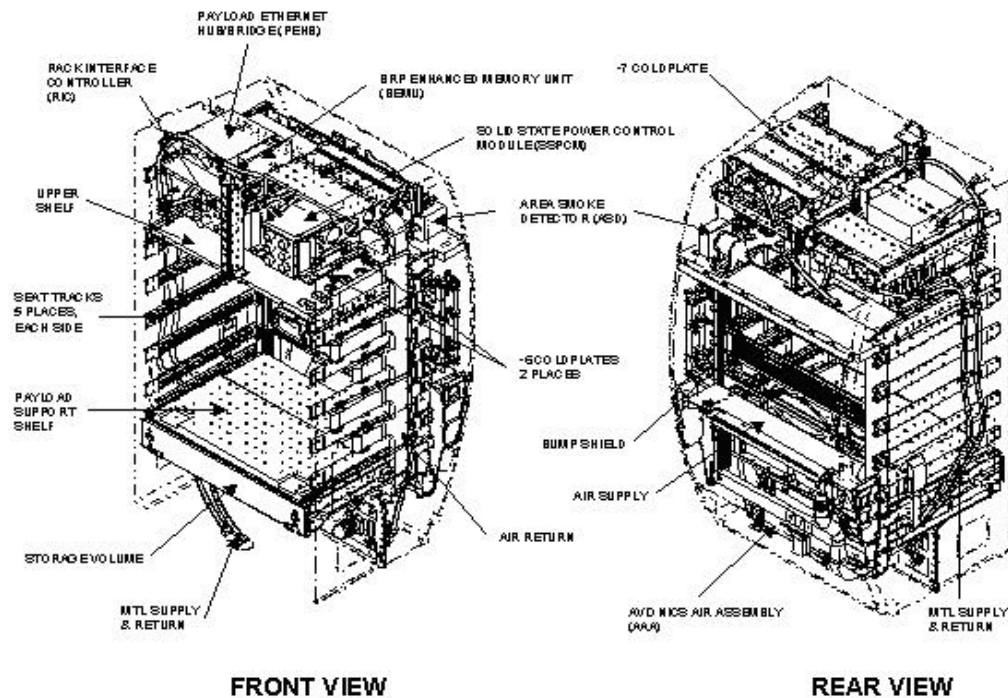
The nadir window in the U.S. Laboratory is a 50.8 cm clear aperture, fused silica window with a total of four panes. The outer most pane, called the debris pane, is 0.86 cm thick and serves as a sacrifice pane to absorb micrometeorite and orbital debris (MMOD) impacts without damaging the pressure panes. This pane is designed to be removable on-orbit, so that any progressive deterioration in the optical quality of the pane can be eliminated with a new pane. The next two panes are the secondary and primary pressure panes. These panes are 3.175 cm thick. The innermost pane, the scratch pane, is designed to protect the primary pressure pane from damage due to loose tools and other debris in the ISS, and will be removable. The combined set of pressure panes and debris pane will have an average optical performance of  $\lambda/14$  wave peak-to-valley over 6" (reference wavelength of 632.8 nm), which will give excellent optical performance. It is estimated that this window will be able to support, without window-induced optical degradation, a payload having a 30.4-cm optical diameter. The windowpanes are given an anti-reflection coating, which provides the best transmittance in the near UV, visible and near IR bands. The window transmittance curve is shown in Figure 6.





**Figure 6. Window Transmittance Curve**

The WOLF rack is based on a modified EXPRESS rack. This rack is designed to give payloads access to power, data, moderate temperature cooling, video downlink, and a stable mounting area with standardized interfaces for payload deployment. The WOLF is designed to handle a payload with maximum dimensions of 53.3 cm wide by 50.8 cm deep by 76.2 cm long and a maximum mass of 136 Kg. The WOLF supplies the interfaces for payload mounts on the sidewalls of the payload volume, and provides a standard bolt interface on the lower surface of the payload volume for payload and avionics mounting. The WOLF and nadir window are shown in Figure 7.



Note: Drawing does not include WOLF Rack front closeout panels, aisle hatch assembly, front connector panel, and the camera connector panels.

**Figure 7. The WOLF and Nadir Window**



Established interfaces within the payload volume provide payloads access to ISS power, data, and cooling utilities. The WOLF will provide payloads with a maximum power of 2 Kw supplied at 28 Vdc. The actual payload allowable power draw will be determined by the thermal system capacity. The power distribution will be 2-14 payload area interfaces, 1 front panel and 1 camera area interface for laptops, and the WOLF avionics systems. Thermal control is provided by forced air-cooling and water-cooling.

The maximum data rate will be up to 8 Mbps. Instrument data will be collected by Ethernet 10BaseT and Serial RS-422 with selectable baud rates, or RS-170A differential video. Instrument data will be distributed based on the data rate. Low rate and health and safety telemetry, medium rate telemetry, and video are available. The interior of the WOLF will be designed to be light tight and low-reflectance, so payloads will be able to observe low-light-level phenomenon such as aurora, and also will be able to support radiometric measurements. At present, design studies are underway to provide the WOLF with passive rack-level vibration isolation so as to provide a stable environment for payload operations.

WOLF Payloads can be operated in any number of ways, ranging from complete crew control of payload operation to completely autonomous, ground-commanded operation with the only crew interaction being initial set-up. For payload set-up, the scratch pane will be removed and a bump shield integral to the WOLF will be deployed to prevent damage to the aft surface of the primary pressure pane by floating debris. Once payloads are mounted and under control, the bump shield will be retracted and the payload optics will be able to be moved into position directly adjacent to the pressure pane to take advantage of the optical quality of the window. It is anticipated that payload developers will provide all necessary mounting hardware, which will, in turn, mount to the interfaces provided in the payload volume.

#### Transport and Installation

The payload developer's flight hardware is transported to the ISS via the shuttle using the ISS Multi-Purpose Logistics Module (MPLM) or via either the STS Mid-deck or SPACEHAB lockers if appropriately sized. Payload installation into the WOLF will be performed by the crew using payload procedures.

### **SECTION 8: PAYLOAD ON-ORBIT OPERATIONS**

The Marshall Space Flight Center's Payload Operations Integration Center (POIC) manages the planning and execution of on-orbit ISS payloads and payload support systems in coordination/unison with distributed International Partner Payload Control Centers, Telescience Support Centers (TSC's) and payload-unique remote facilities. These are supported through various payload operations configurations with the POIC.

One such operations configuration allows users to remain at their home site and obtain an optional copy of the PC Based Telescience Resource Kit (TReK) software and associated Commercial Off the Shelf (COTS) software. TReK software provides local telemetry, command, and database capabilities and access to the POIC capabilities and interfaces. Capabilities provided by TReK include the capability to receive, process, record, and forward real-time and playback telemetry, uplink and update payload commands, perform local exception monitoring, local calculations, word processing, file management, and control telemetry and command processing using local databases. Information needed to populate a TReK database can be downloaded from a supporting facility (POIC or Telescience Support Center) database. Mission execution and mission planning tools can also be accessed from a TReK system.

Additional information on the MSFC POIC and TReK systems can be found on the following website:  
<http://trek.msfc.nasa.gov/>

### **SECTION 9: PAYLOAD SUPPORTED STS/ISS REVIEWS and DELIVERABLES**

The payload developer must support a certain template of ISS Program reviews and deliverables. The following document defines the overall integration template for attached payloads:

ISS Attached Payload: SSP 57057, ISS Payload Integration Template

Payloads using the EXPRESS Pallet, WOLF, JEM-EF and Columbus-EPF sites should refer to site-specific documents for reviews and deliverables.

## SECTION 10: PAYLOAD COST ASSUMPTIONS

Cost Instructions within the AO take precedence over this document. The proposer should not cost shuttle launch or landing services to the extent described below. The NASA ISS Payload Office and KSC budget process currently provides annual funding for KSC Launch and Landing Site payload processing. This processing is based on the Launch Manifest and approved support requirements as developed through the KSC Support requirements data set (SRDS) that is published as part the ISS payloads data set, SSP 52000-PDS.

The existing KSC support capabilities and services that are funded through the NASA budget process are found in SSP 52000-PDS, Payload Data Set Blank Book. The KSC assigned Launch Site Services Manager (LSSM) will assist the Payload Developer (PD) when discussing the nominal levels of support. If facilities, equipment, or service capabilities are requested that are greater than nominal levels of services available, then payload unique requirements, along with the PD unique requirement rationale, will be documented in the Payload Integration Agreement (PIA) Addendum. These unique capabilities or services will be optional costs to the PD unless approved and funded through the ISS Payloads Office. Payload Developers should cost unique capabilities and services in their proposal cost estimate.

Other on-orbit services provided to payloads by the ISS Program include command, telemetry, and data transmission with Level 0 processing performed at the MSFC POIC. Data is then shipped to the user. Limited operations such as high level health and safety monitoring may also be performed at the MSFC POIC. Operating systems such as the MSFC TReK, for command and telemetry from a payload operations center, can be provided at a minimal cost to the payload developer and must be included in the cost estimate. This assumes the operating platform is provided by the payload developer.

### Full Truss Site Payload

Since a Full Truss payload is usually a unique payload attached to the ISS truss, the payload developer is generally responsible for providing all attach mounting hardware, all necessary grapple fixtures, and interfaces for both the ISS S3 site as well as installation and mounting hardware in the shuttle. Payload mockups will most likely be required for crew training and must be provided by the payload developer. ISS Program-furnished flight equipment at no cost can be found in SSP 57061, Standard Payload Integration Agreement for Unpressurized Payloads, such as, one Berthing Camera System, one Power Video Grapple Fixture and possibly some EVA tools and aids. Payload Developer procured flight equipment from NASA is also defined in the above SSP 57061 reference and includes items such as the Passive Umbilical Mechanism Assembly and any Flight Releasable or Payload Data Grapple Fixtures.

### EXPRESS Pallet Payload/Columbus EPF Payload

The ISS Program will provide the EXPRESS Pallet adapter hardware and the payload shipping container to EXPRESS Pallet payloads and any NASA payloads planning to use the Columbus EPF as defined in SSP 57061. The integration of an EXPRESS Pallet payload to the Pallet is provided without charge, but must have personnel support by the payload developer. Payload mockups for crew training are not anticipated to be required unless the payload exceeds the standard volume. However, if mockups are required, the payload developer must provide these. Interfaces to the EXPRESS Pallet or the Columbus EPF are the joint responsibility of the payload developer and the ISS facility project. Interfaces between the EXPRESS Pallet or the Columbus EPF and the ISS and between the EXPRESS Pallet and the shuttle, including the associated documentation, are the responsibility for the EXPRESS Pallet project/Columbus EPF project and are provided at no cost.

### Japanese Experiment Module – Exposed Facility

Payload developers planning to attach to the JEM-EF will be provided one JEM-EF Payload Interface Unit (PIU) at no cost as defined in SSP 57061. However, the payload developer must provide the required grapple fixtures. At present, it is not clear if a Flight Releasable Attach Mechanism (FRAM) will be provided at no cost. It is anticipated that a U.S. payload will most likely use a NASA provided Universal Logistics Carrier (ULC) to transport the payload to the ISS. This service and associated interfaces are provided at no cost to the payload developer. Transportation to the ISS may also be available on the Japanese HTV vehicle; however, the return flight is required via the shuttle. Interfaces to the JEM-EF are the joint responsibility of the payload developer and the JEM-EF facility provider. Interfaces between the JEM-EF and the ISS are the responsibility of the JEM-EF provider and are no cost to the payload developer. Currently, issues exist for transport and installation of JEM-EF payloads >500 Kg.

Payload mockups for crew training are not anticipated to be required unless the payload exceeds the standard volume. However, if mockups are required, the payload developer must provide these.

## Window Observational Research Facility

There are no anticipated unique facility use costs for the WORF. Payload mockups for crew training are not anticipated to be required unless the payload involves unusual complexity or unique crew activity. Interfaces to the WORF are the joint responsibility of the payload developer and the MSFC WORF Project. Interfaces between the WORF and the ISS, including associated documentation, are the responsibility of the MSFC WORF Project and are provided at no cost.

### **SECTION 11: APPLICABLE DOCUMENTATION**

NASA applicable documents are available through website libraries maintained by the ISS Program at the Johnson Space Center in Houston, Texas. JEM-EF and Columbus EPF documents are maintained by NASDA and ESA, respectively. Documents may be obtained by requesting online access to the JSC Program Automated Library System by completing the attached online access request form and returning it to [richard.c.wadle@nasa.gov](mailto:richard.c.wadle@nasa.gov). Please contact Richard Wadle at 281-244-7389 with any questions. The OES/OSS RPO web site at <http://rpo-iss.gsfc.nasa.gov/> contains general information on ISS attached payload accommodations.

#### General

ISS User's Guide-Release 2.0 (ISSUG), <http://spaceflight.nasa.gov/station/reference/index.html>

SSP 50431, Space Station Program Requirements for Payloads

SSP 57061, Standard Payload Integration Agreement for Unpressurized Payloads

#### Full Truss Payloads

SSP 57021, Attached Payloads Accommodation Handbook

SSP 57003, Attached Payloads Interface Requirements Document

SSP 57013, Generic Attached Payloads Verification Plan

SSP 52000-PIA-UPP, Payload Integration Agreement Blank Book for Unpressurized Payloads

#### EXPRESS Pallet Payloads

SSP 52000-IDD-EPP, EXPRESS Pallet Interface Definition Document, Draft 5

SSP 52000-PAH-EPP, Payload Accommodations Handbook EXPRESS Pallet Payloads, Draft 3

SSP 52000-PVP-EPP, Generic Payload Verification Plan EXPRESS Pallet Payloads, Draft 3

#### JEM-EF Payloads

JBX-98079, Introductory Guidebook for JEM Exposed Facility Potential Users

NASDA-ESPC-2563 JEM Payload Accommodation Handbook

#### EPF Payloads

COL-RIBRE-MA-00007-00, Columbus Payload Accommodation Handbook

#### WORF Payloads

SSP 52000-PIH-WRP, WORF Rack Payload Integration Handbook, Baseline, Volumes 1-6

#### ISS Mission Template

SSP 57057 ISS Payload Integration Template

### **SECTION 12: OES/OSS RESEARCH PROGRAM OFFICE FOR ISS UTILIZATION**

The OES/OSS Research Program Office (RPO) for ISS utilization is the primary point of contact for earth and space science proposers interested in flying NASA payloads on the ISS. Questions regarding ISS utilization, accommodations, and interfaces should be addressed to the Research Program Manager, GSFC Code 804.G, Greenbelt, MD 20771, (301) 286-9833, fax (301) 286-1694, [steven.c.dunker@nasa.gov](mailto:steven.c.dunker@nasa.gov).

The OES/OSS RPO assists Headquarters with strategic and tactical planning, international bartering, and serves as the OES/OSS representative on ISS review panels, task committees and working groups. The OES/OSS RPO also serves as a payload representative to the ISS Program Office for review of ISS documentation pertaining to payloads and for providing Increment Scientists to represent the OES/OSS payloads during on-orbit mission operations. The OES/OSS RPO is responsible for working ISS allocations, manifesting, ISS and STS interfaces for payloads and issues regarding the above. Mission management of OES/OSS ISS payloads will be handled in the same manner as other Explorer or Earth Probes missions, however, the OES/OSS RPO will support the selected ISS payload in coordinating payload planning, accommodations, allocations, manifesting, development, integration, operations and payload retrieval. The OES/OSS RPO acts to shepherd the payload developers through the STS/ISS systems, documentation, procedures, and reviews from beginning to end. The OES/OSS RPO has final Certificate of Flight Readiness signature responsibility to the ISS Payloads Office.

## ISS Online Access READ-ONLY Request Form

Make sure you fill this form completely!

### User Information:

NAME:

CITIZENSHIP: US: YES \_\_\_ NO \_\_\_ If NO indicate country of Citizenship:

E-MAIL ADDRESS:

ORGANIZATION:

PHONE NUMBER:

CONTRACTOR NUMBER:

IP NUMBER (Static):

(If you **DO NOT** have a static IP number please describe how you are accessing the internet. This information must include whether you are accessing via a proxy server or firewall and their IP numbers)

### NASA Point of Contact Information:

NAME:

PHONE NUMBER:

LOCATION (SITE/BLDG/ROOM):

REASON ACCESS IS REQUIRED :

### IMPORTANT!

Remember to fill in this form in full. This is important so that your request can be processed as soon as possible. Failure to do so may delay or inhibit the satisfaction of this request. If you have any questions, please contact Richard Wadle at 281-244-7389.

After completing this form please email it to [richard.c.wadle@nasa.gov](mailto:richard.c.wadle@nasa.gov).

PALS website: <http://iss-www.jsc.nasa.gov/ss/issapt/pals/>

PALS Administrator (Connie Wright, 281-461-2353, [connie.m.wright1@jsc.nasa.gov](mailto:connie.m.wright1@jsc.nasa.gov))